

ABSTRACT OF THE DISCLOSURE

In the blade structure in a gas turbine, front-edge including angles are made large. As a result, a curve of a relative relationship between incidence angles α_1 and β_1 and pressure loss becomes mild. Entrance metal angles are made small. As a result, it becomes possible to make the incidence angles small. Chord length of a chip portion of a moving blade is made large. As a result, it becomes possible to make small the deceleration on a rear surface of the chip portion of the moving blade. Accordingly, it becomes possible to make the pressure loss small, and therefore, it becomes possible to improve the turbine efficiency.

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